SECTION 5

PERFORMANCE

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5.1 - ACOUSTIC LIMITATION

	Maximum noise level permissible	Demonstrated noise level
FAR PART 36, Appendix G - Amdt 17	**85 dB(A)**	**77.4 dB(A)**
OACI, Annex 16, Chapter 10, Appendix 6	**88 dB(A)**	**80.4 dB(A)**

TBM 700 airplane has received the noise limitation type certificate Nr N181 dated 31st January 1990.

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5.2 - AIRSPEED CALIBRATION

NOTE :
Indicated airspeeds (IAS) : instrument error supposed to be null (power configuration for cruise condition flight).

FLAPS UP LDG GR UP		FLAPS TO LDG GR DN		FLAPS LDG LDG GR DN	
KIAS	KCAS	KIAS	KCAS	KIAS	KCAS
125	127	70	69	60	58
150	152	80	80	70	68
175	177	90	90	80	78
200	203	100	101	90	88
225	228	120	121	100	98
250	253	140	141	110	108
266	270	160	162	120	118
MPH IAS	MPH CAS	MPH IAS	MPH CAS	MPH IAS	MPH CAS
144	146	81	79	69	67
173	175	92	92	81	78
201	204	104	104	92	90
230	233	115	116	104	101
259	262	138	139	115	113
288	292	161	162	127	124
307	311	184	187	138	136

Figure 5.2.1 - NORMAL STATIC SOURCE

FLAPS UP LDG GR UP		FLAPS TO LDG GR DN		FLAPS LDG LDG GR DN	
KIAS	KCAS	KIAS	KCAS	KIAS	KCAS
125	124	70	70	60	59
150	149	80	80	70	69
175	174	90	90	80	79
200	199	100	100	90	90
225	224	120	120	100	100
250	249	140	139	110	110
271	270	160	159	120	120
MPH IAS	MPH CAS	MPH IAS	MPH CAS	MPH IAS	MPH CAS
144	142	81	81	69	68
173	171	92	92	81	79
201	200	104	104	92	91
230	229	115	115	104	104
259	258	138	138	115	115
288	287	161	160	127	127
312	311	184	183	138	138

Figure 5.2.2 - ALTERNATE STATIC SOURCE (BLEED ON)

5.3 - CABIN PRESSURIZATION ENVELOPE

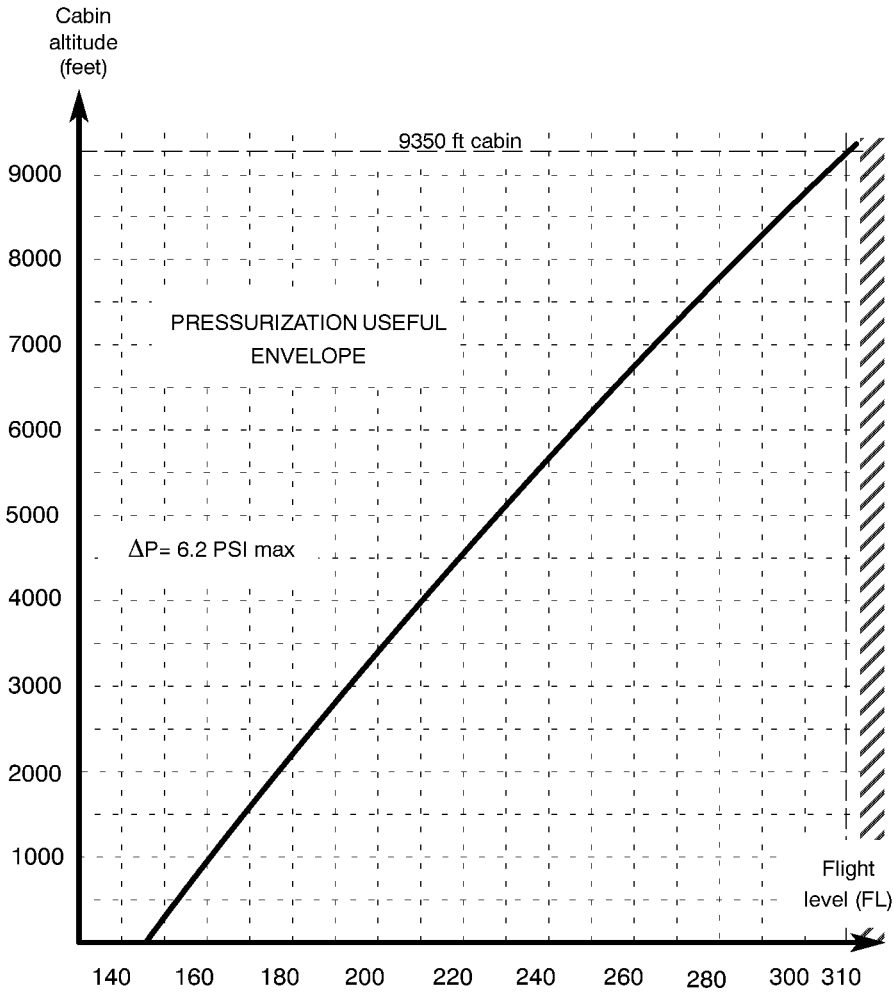

Figure 5.3.1 - CABIN PRESSURIZATION ENVELOPE

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5.4 - SAT - IOAT CONVERSIONS

NOTE :
These indicated temperatures are available for stabilized cruise at normal operating power.

Pressure altitude (feet)	ISA - 20°C		ISA - 10°C		ISA		ISA + 10°C		ISA + 20°C	
	SAT	IOAT	SAT	IOAT	SAT	IOAT	SAT	IOAT	SAT	IOAT
SL	- 05	02	05	12	15	22	25	32	35	42
2000	- 09	- 02	01	08	11	18	21	28	31	38
4000	- 13	- 06	- 03	04	07	14	17	25	27	35
6000	- 17	- 10	- 07	00	03	11	13	21	23	31
8000	- 21	- 13	- 11	- 03	- 01	07	09	17	19	27
10000	- 25	- 17	- 15	- 07	- 05	03	05	13	15	23
12000	- 29	- 21	- 19	- 11	- 09	- 01	01	10	11	20
14000	- 33	- 25	- 23	- 14	- 13	- 04	- 03	06	07	16
16000	- 37	- 28	- 27	- 18	- 17	- 08	- 07	02	03	12
18000	- 41	- 32	- 31	- 22	- 21	- 12	- 11	- 01	- 01	08
20000	- 45	- 36	- 35	- 26	- 25	- 15	- 15	- 05	- 05	04
22000	- 48	- 39	- 38	- 29	- 28	- 19	- 18	- 09	- 08	00
24000	- 52	- 43	- 42	- 33	- 32	- 23	- 22	- 13	- 12	- 04
26000	- 56	- 47	- 46	- 36	- 36	- 27	- 26	- 17	- 16	- 08
28000	- 60	- 50	- 50	- 40	- 40	- 31	- 30	- 21	- 20	- 12
30000	- 64	- 54	- 54	- 45	- 44	- 35	- 34	- 26	- 24	- 16

Figure 5.4.1 - SAT - IOAT CONVERSIONS

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5.5 - STALL SPEEDS

AIR-PLANE WEIGHT	CONFIG.		BANK											
	FLIGHT IDLE		0°			30°			45°			60°		
	LDG GR	Flaps	KIAS	KCAS	MPH IAS	KIAS	KCAS	MPH IAS	KIAS	KCAS	MPH IAS	KIAS	KCAS	MPH IAS
4850 lbs (2200 kg)	UP	UP	65	66	75	70	71	81	78	79	90	91	93	105
	DN	TO	62	63	71	67	68	77	73	75	84	87	89	100
	DN	LDG	53	53	61	57	57	66	63	63	73	75	75	86
5512 lbs (2500 kg)	UP	UP	70	71	81	75	76	86	82	84	94	98	100	113
	DN	TO	66	67	76	71	72	82	78	80	90	93	95	107
	DN	LDG	57	57	66	61	61	70	68	68	78	81	81	93
6579 lbs (2984 kg)	UP	UP	75	76	86	80	82	92	88	90	101	105	107	121
	DN	TO	71	72	82	75	77	86	84	86	97	100	102	115
	DN	LDG	61	61	70	66	66	76	73	73	84	86	86	99

Figure 5.5.1 - STALL SPEEDS

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5.6 - WIND COMPONENTS

EXAMPLE : Angle between wind direction and flight path : 50 °
Headwind : 8 kts
Crosswind : 10 kts
Wind speed : 13 kts

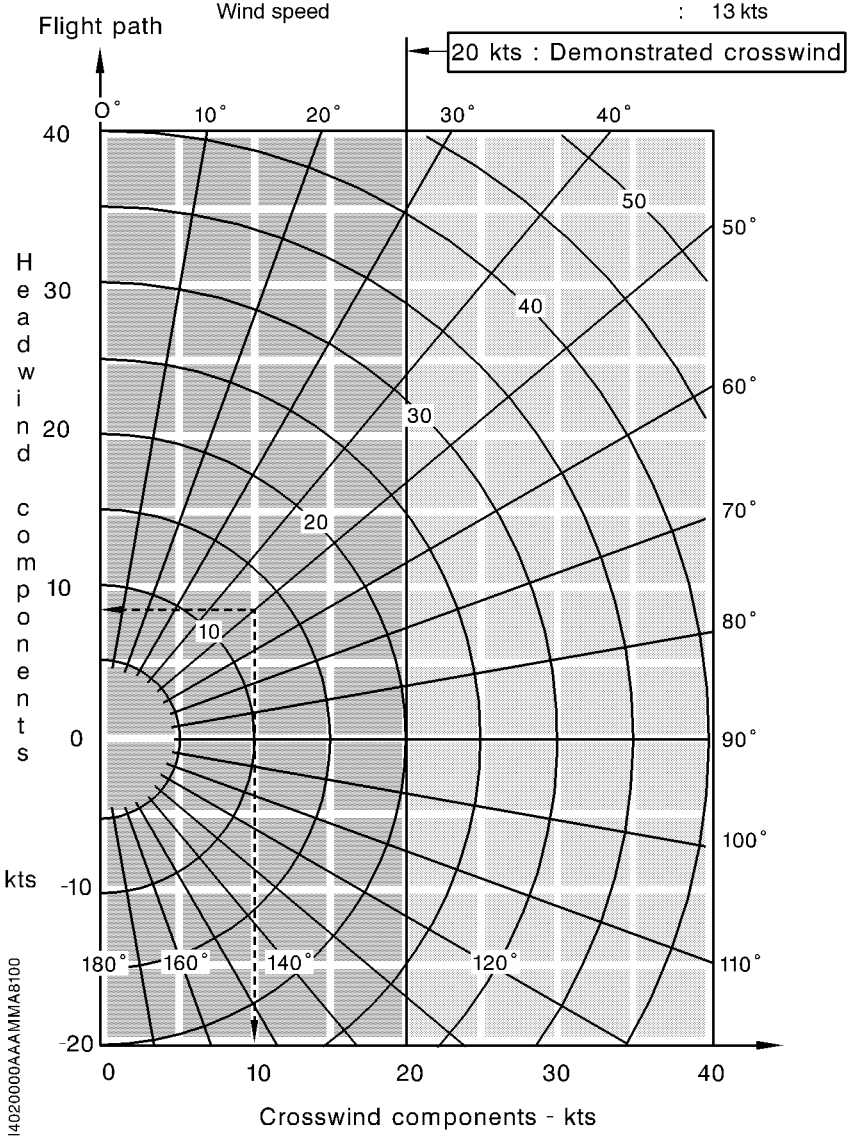

Figure 5.6.1 - WIND COMPONENTS

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5.7 - ENGINE OPERATION

The following tables must be used during normal operation of the airplane.

The following conditions are given :

- Np = 2000 RPM,
- BLEED ON.

The torque must be set at or below the value corresponding to the local conditions of flight level and temperature.

Example : for FL = 260 and IOAT = - 25°C, the following tables give the maximum torque to be set.

Maximum climb power : TRQ = 84 % for IAS = 130 KIAS
(Add 1 % of TRQ for each additional 15 KIAS on climb airspeed)
(cf. tables Figures 5.7.1A and 5.7.1B)

Maximum cruise power : TRQ = 98 %
(cf. tables Figures 5.7.2A and 5.7.2B)

Recommended cruise power : TRQ = 92 %
(cf. tables Figures 5.7.3A and 5.7.3B)

CAUTION

THE TRQ SETTING MUST NEVER EXCEED 100 % FOR NP = 2000 RPM

REMARK :
The engine ITT limit at 800°C during continuous operation may be used in case of operational need. However, in order to ensure a good engine aging, an ITT limit at 785°C is recommended during continuous operation (climb and cruise).

ENGINE OPERATION

Conditions : ***Maximum climb power (FL ≤ 200)*** ☐ ISA
Landing gear and flaps UP
IAS = 130 KIAS - Np = 2000 RPM - BLEED ON

NOTE :
Add 1 % of TRQ for each additional 10 KCAS on climb airspeed

T° (°C)		FLIGHT LEVEL (FL)										
SAT	IOAT	100	110	120	130	140	150	160	170	180	190	200
- 25	- 19											
- 23	- 17											
- 21	- 15											
- 19	- 13		**TRQ = 100 %**									100
- 17	- 11											99
- 15	- 09											97
- 13	- 07										100	95
- 11	- 05										98	93
- 09	- 03									100	96	91
- 07	- 01									99	94	89
- 05	+ 01								100	97	92	87
- 03	+ 03								99	95	90	85
- 01	+ 05								97	93	87	82
+ 01	+ 07							100	95	90	85	80
+ 03	+ 09							98	93	88	82	78
+ 05	+ 11						100	95	91	86	80	75
+ 07	+ 13						98	93	88	83	78	72
+ 09	+ 15					100	95	90	86	81	74	69
+ 11	+ 17				100	97	93	88	83	78	71	66
+ 13	+ 19				99	95	90	85	80	74	69	
+ 15	+ 21			100	97	92	88	82	77	71		
+ 17	+ 23			99	94	90	84	79	67			
+ 19	+ 25		100	96	92	86	81	67				
+ 21	+ 27		98	94	88	83	68					
+ 23	+ 29	100	96	90	76	67						
+ 25	+ 31	97	84	76	67							
+ 27	+ 33	84	75	67								
+ 29	+ 35	74	68									
+ 31	+ 37	69										

CAUTION

THE TRQ SETTING MUST NEVER EXCEED 100 % FOR Np = 2000 RPM

Figure 5.7.1A - ENGINE OPERATION
[Maximum climb power (FL ≤ 200)]

ENGINE OPERATION

Conditions : ***Maximum climb power (FL ≥ 200)*** ☐ ISA
Landing gear and flaps UP
IAS = 130 KIAS - Np = 2000 RPM - BLEED ON

NOTE :
Add 1 % of TRQ for each additional 10 KCAS on climb airspeed

T° (°C)		FLIGHT LEVEL (FL)											
SAT	IOAT	200	210	220	230	240	250	260	270	280	290	300	310
- 67	- 60												91
- 65	- 58										100	94	89
- 63	- 56										97	92	86
- 61	- 54									100	95	90	85
- 59	- 52									99	93	88	83
- 57	- 50									97	91	85	81
- 55	- 48		**TRQ = 100 %**						100	94	89	84	79
- 53	- 46								98	92	87	82	77
- 51	- 44							100	95	90	85	80	75
- 49	- 42							99	93	88	83	79	74
- 47	- 40							97	92	87	82	77	73
- 45	- 38						100	95	90	85	80	76	72
- 43	- 36						98	93	88	83	79	75	70
- 41	- 34						97	92	87	82	78	73	69
- 39	- 32					100	95	90	85	81	76	72	67
- 37	- 30					98	94	89	84	79	74	70	65
- 35	- 28					97	92	87	82	77	73	68	64
- 33	- 26				100	95	91	86	81	76	71	66	62
- 31	- 25				99	94	89	84	79	74	69	65	61
- 29	- 23				97	92	87	82	77	72	68	63	59
- 27	- 21			100	95	90	85	80	75	70	66	61	57
- 25	- 19			98	93	88	83	78	73	69	64	59	55
- 23	- 17		100	95	91	86	81	76	71	67	62	58	54
- 21	- 15		98	94	89	84	79	74	70	65	60	57	53
- 19	- 13	100	96	92	87	82	77	73	68	63	59	55	
- 17	- 11	99	94	90	85	80	75	71	66	61	57	53	
- 15	- 09	97	92	88	83	78	74	69	64	59	55		
- 13	- 07	95	90	86	81	76	72	67	62	57	53		
- 11	- 05	93	88	84	79	74	70	65	60	55			
- 09	- 03	91	86	82	77	72	68	63	58	53			
- 07	- 01	89	84	80	75	70	65	60	56				
- 05	+ 01	87	82	77	73	68	63	58	54				
- 03	+ 03	85	80	75	70	66	60	55					
- 01	+ 05	82	78	73	68	63	57	53					
+ 01	+ 07	80	75	70	65	60	55						
+ 03	+ 09	78	73	67	62	57							
+ 05	+ 11	75	70	65	59								
+ 07	+ 13	72	67	62									
+ 09	+ 15	69	64										
+ 11	+ 17	66											

CAUTION
THE TRQ SETTING MUST NEVER EXCEED 100 % FOR Np = 2000 RPM

Figure 5.7.1B - ENGINE OPERATION
[Maximum climb power (FL ≥ 200)]

ENGINE OPERATION

Conditions : ***Maximum cruise power (FL ≤ 200)*** ☐ ISA
Landing gear and flaps UP
Np = 2000 RPM - BLEED ON

NOTE :
Use preferably recommended cruise power

T° (°C)		FLIGHT LEVEL (FL)										
SAT	IOAT	100	110	120	130	140	150	160	170	180	190	200
- 25	- 15											
- 23	- 13											
- 21	- 11											
- 19	- 09		**TRQ = 100 %**									
- 17	- 07											
- 15	- 05											
- 13	- 03											
- 11	- 02											
- 09	00											100
- 07	+ 02											99
- 05	+ 04											97
- 03	+ 06										100	95
- 01	+ 08										98	93
+ 01	+ 10									100	95	91
+ 03	+ 12									98	93	88
+ 05	+ 14								100	96	91	85
+ 07	+ 16								98	93	88	82
+ 09	+ 18							100	96	91	84	78
+ 11	+ 20							98	93	87	81	75
+ 13	+ 22							97	91	85	79	
+ 15	+ 24						100	93	87	82		
+ 17	+ 26					100	96	90	84			
+ 19	+ 28					98	92	86				
+ 21	+ 29				100	95	89					
+ 23	+ 31			100	97	84						
+ 25	+ 33		100	92	84							
+ 27	+ 35	100	92	83								
+ 29	+ 37	91	83									
+ 31	+ 39	82										

CAUTION

THE TRQ SETTING MUST NEVER EXCEED 100 % FOR Np = 2000 RPM

Figure 5.7.2A - ENGINE OPERATION
[Maximum cruise power (FL ≤ 200)]

ENGINE OPERATION

Conditions : ***Maximum cruise power (FL ≥ 200)*** ☐ ISA
Landing gear and flaps UP
Np = 2000 RPM - BLEED ON

NOTE :
Use preferably recommended cruise power

T° (°C)		FLIGHT LEVEL (FL)											
SAT	IOAT	200	210	220	230	240	250	260	270	280	290	300	310
- 67	- 56												100
- 65	- 54												99
- 63	- 52												97
- 61	- 50											100	95
- 59	- 48											98	93
- 57	- 46										100	96	91
- 55	- 44										98	94	89
- 53	- 42			**TRQ = 100 %**							97	92	87
- 51	- 40									100	95	90	85
- 49	- 38									98	93	88	83
- 47	- 36									97	91	86	81
- 45	- 34								100	95	89	84	79
- 43	- 32								99	93	88	82	77
- 41	- 30								97	91	86	80	75
- 39	- 28							100	95	90	84	79	74
- 37	- 26							99	93	88	82	77	72
- 35	- 24							97	91	86	81	75	70
- 33	- 22						100	95	89	84	79	73	68
- 31	- 20						98	93	88	82	77	71	66
- 29	- 19						96	91	86	80	75	69	64
- 27	- 17					100	94	89	84	78	73	67	62
- 25	- 15					98	92	87	82	76	71	66	60
- 23	- 13				100	96	90	85	80	74	69	64	59
- 21	- 11				98	94	88	83	78	72	67	62	57
- 19	- 09				97	92	86	81	76	70	65	61	
- 17	- 07			100	95	90	84	79	74	68	63	59	
- 15	- 05			98	93	88	82	77	72	66	61	57	
- 13	- 03		100	96	91	86	80	74	69	64	59	55	
- 11	- 02		98	94	88	84	78	72	67	62	57	53	
- 09	00	100	96	92	86	81	75	70	65	59	54	51	
- 07	+ 02	99	94	89	84	79	73	67	62	56	52		
- 05	+ 04	97	92	87	81	77	71	65	59	54			
- 03	+ 06	95	90	85	79	74	68	62	56				
- 01	+ 08	93	87	82	76	71	65	60					
+ 01	+ 10	91	85	80	73	69	62						
+ 03	+ 12	88	82	77	71	66							
+ 05	+ 14	85	79	74	68								
+ 07	+ 16	82	76	72									
+ 09	+ 18	78	73										
+ 11	+ 20	75											

CAUTION
THE TRQ SETTING MUST NEVER EXCEED 100 % FOR Np = 2000 RPM

Figure 5.7.2B - ENGINE OPERATION
[Maximum cruise power (FL ≥ 200)]

ENGINE OPERATION

Conditions : ***Normal (recommended) cruise power (FL ≤ 200)*** ☐ ISA
Landing gear and flaps UP
Np = 2000 RPM - BLEED ON

T° (°C)		FLIGHT LEVEL (FL)										
SAT	IOAT	100	110	120	130	140	150	160	170	180	190	200
- 25	- 15											
- 23	- 13											
- 21	- 11											
- 19	- 09		TRQ = 100 %									
- 17	- 07											
- 15	- 05											100
- 13	- 03											98
- 11	- 02											96
- 09	00										100	94
- 07	+ 02										98	92
- 05	+ 04									100	96	90
- 03	+ 06									99	93	87
- 01	+ 08								100	96	91	85
+ 01	+ 10								98	93	88	82
+ 03	+ 12							100	95	90	85	79
+ 05	+ 14							98	93	88	83	76
+ 07	+ 16						100	95	90	85	81	74
+ 09	+ 18						98	93	88	83	78	71
+ 11	+ 20					100	95	90	86	80	74	67
+ 13	+ 22					98	93	88	82	76	62	
+ 15	+ 24				100	95	91	85	72	63		
+ 17	+ 26			100	97	93	84	72	63			
+ 19	+ 28		100	99	95	81	72	64				
+ 21	+ 29	100	94	88	80	72	64					
+ 23	+ 31	93	88	80	72	65						
+ 25	+ 33	87	79	71	66							
+ 27	+ 35	79	73	67								
+ 29	+ 37	74	69									
+ 31	+ 39	70										

CAUTION

THE TRQ MUST NEVER EXCEED 100 % FOR Np = 2000 RPM

Figure 5.7.3A - ENGINE OPERATION
[Normal (recommended) cruise power (FL ≤ 200)]

ENGINE OPERATION

Conditions : ***Normal (recommended) cruise power (FL ≥ 200)*** ☐ ISA
Landing gear and flaps UP
Np = 2000 RPM - BLEED ON

T° (°C)		FLIGHT LEVEL (FL)											
SAT	IOAT	200	210	220	230	240	250	260	270	280	290	300	310
- 67	- 56												95
- 65	- 54											100	93
- 63	- 52											98	91
- 61	- 50										100	95	89
- 59	- 48										98	93	87
- 57	- 46										96	90	85
- 55	- 44									100	94	88	83
- 53	- 42									98	92	86	81
- 51	- 41								100	96	90	84	79
- 49	- 39								99	94	88	82	77
- 47	- 37								97	92	86	80	75
- 45	- 35							100	95	90	85	78	73
- 43	- 33							99	94	88	83	76	71
- 41	- 31		TRQ = 100 %					97	92	86	81	75	70
- 39	- 29						100	95	90	84	79	74	69
- 37	- 27						98	94	87	82	77	72	67
- 35	- 25						97	92	85	80	75	69	65
- 33	- 23					100	95	90	83	78	73	67	63
- 31	- 21					98	93	88	81	76	71	66	61
- 29	- 19				100	96	91	86	79	74	69	64	59
- 27	- 17				98	94	89	83	77	72	67	62	57
- 25	- 15				96	92	87	81	75	70	65	60	56
- 23	- 13			100	94	90	85	79	73	68	63	58	54
- 21	- 12			98	92	87	83	76	71	66	61	57	52
- 19	- 10		100	96	90	85	80	74	69	64	59	55	
- 17	- 08		98	93	88	83	78	72	67	62	58	53	
- 15	- 06	100	96	91	86	81	76	70	65	60	56	51	
- 13	- 04	98	94	89	83	78	73	67	63	58	54	50	
- 11	- 02	96	92	87	81	76	71	65	60	56	52	48	
- 09	00	94	89	84	78	74	69	63	58	54	50	46	
- 07	+ 02	92	87	82	76	71	66	60	56	52	48		
- 05	+ 04	90	84	79	74	68	64	58	54	50			
- 03	+ 06	87	82	77	71	66	61	56	52				
- 01	+ 08	85	79	74	68	63	58	53					
+ 01	+ 10	82	77	72	66	60	55						
+ 03	+ 12	79	74	69	63	58							
+ 05	+ 14	76	71	67	60								
+ 07	+ 16	74	68	64									
+ 09	+ 18	71	65										
+ 11	+ 20	67											

CAUTION
THE TRQ MUST NEVER EXCEED 100 % FOR Np = 2000 RPM

Figure 5.7.3B - ENGINE OPERATION
[Normal (recommended) cruise power (FL ≥ 200)]

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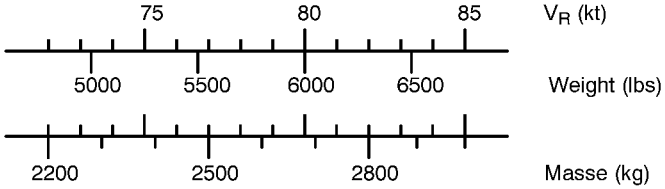
5.8 - TAKEOFF DISTANCES

WEIGHT : 5512 lbs (2500 kg)

Associated conditions :
- Landing gear DN and flaps TO
- 15° of attitude - TRQ = 100 %
- Np = 2000 RPM - BLEED ON
- Hard, dry and level runway
- GR = Ground roll (in ft)
- D_{50} = Takeoff distance (clear to 50 ft) (in ft)
- Rotation speed choice (V_R)



WEIGHT : 5512 lbs (2500 kg) At 50 ft = 91 KIAS - 105 MPH IAS								
PRESSURE ALTITUDE ft	ISA - 35°C		ISA - 20°C		ISA - 10°C		ISA	
	GR	D50	GR	D50	GR	D50	GR	D50
0	787	1280	886	1411	951	1493	1017	1591
2000	886	1411	984	1558	1066	1657	1132	1772
4000	984	1558	1099	1722	1181	1837	1280	1968
6000	1099	1722	1230	1903	1329	2051	1444	2215
8000	1230	1903	1394	2149	1526	2329	1657	2510
PRESSURE ALTITUDE ft	ISA + 10°C		ISA + 20°C		ISA + 30°C		ISA + 37°C	
	GR	D50	GR	D50	GR	D50	GR	D50
0	1083	1690	1148	1788	1214	1903	1247	1969
2000	1214	1870	1296	1985	1378	2133	1444	2231
4000	1363	2100	1476	2247	1575	2411	1640	2526
6000	1575	2379	1690	2559	1837	2756	1919	2887
8000	1804	2707	1968	2920	2100	3133	2198	3281

Figure 5.8.1 - TAKEOFF DISTANCES - 5512 lbs (2500 kg)

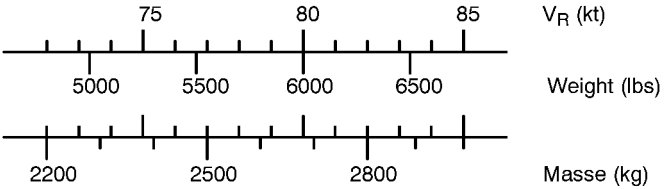
Corrections :
- Reduce total distances of 10 % every 10 kts of headwind
- Increase total distances of 30 % every 10 kts of rear wind
- Increase by : 7 % on hard sod; 25 % on high grass; 10 % on short grass; 30 % on slippery runway; 15 % on wet runway

NOTE :
Between ISA + 30°C and ISA + 37°C, it may be necessary to cut-off the Bleed in order to set TRQ = 100 % during takeoff while respecting the engine limitations. In this case, reduce power after takeoff to set the Bleed ON.

TAKEOFF DISTANCES

WEIGHT : 6579 lbs (2984 kg)

Associated conditions :
- Landing gear DN and flaps TO
- 15° of attitude - TRQ = 100 %
- Np = 2000 RPM - BLEED ON
- Hard, dry and level runway
- GR = Ground roll (in ft)
- D_{50} = Takeoff distance (clear to 50 ft) (in ft)
- Rotation speed choice (V_R)



WEIGHT : 6579 lbs (2984 kg)	At 50 ft = 94 KIAS - 108 MPH IAS							
PRESSURE ALTITUDE ft	ISA - 35°C		ISA - 20°C		ISA - 10°C		ISA	
	GR	D50	GR	D50	GR	D50	GR	D50
0	1083	1673	1214	1870	1280	2001	1378	2133
2000	1214	1870	1345	2067	1444	2198	1542	2362
4000	1345	2067	1509	2297	1640	2461	1739	2625
6000	1509	2297	1706	2559	1837	2723	1968	2920
8000	1706	2559	1903	2854	2067	3051	2231	3281
PRESSURE ALTITUDE ft	ISA + 10°C		ISA + 20°C		ISA + 30°C		ISA + 37°C	
	GR	D50	GR	D50	GR	D50	GR	D50
0	1476	2264	1575	2395	1690	2559	1755	2657
2000	1673	2493	1772	2657	1903	2854	1969	2953
4000	1870	2789	2001	2953	2149	3182	2231	3314
6000	2100	3117	2297	3346	2461	3609	2543	3740
8000	2428	3543	2657	3839	2854	4134	2969	4298

Figure 5.8.2 - TAKEOFF DISTANCES - 6579 lbs (2984 kg)

Corrections :
- Reduce total distances of 10 % every 10 kts of headwind
- Increase total distances of 30 % every 10 kts of rear wind
- Increase by : 7 % on hard sod; 25 % on high grass; 10 % on short grass; 30 % on slippery runway; 15 % on wet runway

NOTE :
Between ISA + 30°C and ISA + 37°C, it may be necessary to cut-off the Bleed in order to set TRQ = 100 % during takeoff while respecting the engine limitations. In this case, reduce power after takeoff to set the Bleed ON.

5.9 - CLIMB PERFORMANCE

CLIMB SPEEDS (IAS = 130 KIAS)

Conditions : Maximum climb power
Landing gear and flaps UP
IAS = 130 KIAS - BLEED ON

Airplane weight	Pressure altitude (feet)	RATE OF CLIMB (ft/min)					
		ISA - 20°C	ISA - 10°C	ISA	ISA + 10°C	ISA + 20°C	ISA + 30°C
4850 lbs (2200 kg)	SL	3060	2920	2800	2690	2590	2480
	2000	3030	2890	2770	2660	2550	2455
	4000	3000	2860	2740	2630	2520	2415
	6000	2970	2830	2700	2590	2480	2380
	8000	2930	2800	2660	2550	2440	2340
5512 lbs (2500 kg)	SL	2600	2485	2380	2285	2190	2105
	2000	2570	2455	2350	2250	2160	2080
	4000	2540	2425	2320	2220	2130	2045
	6000	2510	2395	2290	2185	2090	2010
	8000	2480	2350	2250	2150	2035	1975
6579 lbs (2984 kg)	SL	2050	1955	1875	1795	1720	1640
	2000	2025	1925	1840	1765	1690	1620
	4000	1995	1900	1815	1735	1660	1585
	6000	1970	1870	1780	1700	1625	1555
	8000	1935	1840	1745	1665	1590	1520

Figure 5.9.1 - CLIMB SPEEDS (IAS = 130 KIAS)

CLIMB PERFORMANCE

CLIMB SPEEDS (IAS = 160 KIAS)

Conditions : Maximum climb power
Landing gear and flaps UP
IAS = 160 KIAS - BLEED ON

Airplane weight	Pressure altitude (feet)	RATE OF CLIMB (ft/min)					
		ISA - 20°C	ISA - 10°C	ISA	ISA + 10°C	ISA + 20°C	ISA + 30°C
4850 lbs (2200 kg)	SL	2680	2560	2440	2330	2220	2120
	2000	2640	2500	2390	2280	2180	2080
	4000	2590	2460	2340	2230	2130	2030
	6000	2550	2420	2290	2180	2080	1980
	8000	2500	2360	2240	2130	2030	1925
5512 lbs (2500 kg)	SL	2290	2180	2000	1980	1890	1805
	2000	2250	2135	2030	1940	1850	1765
	4000	2200	2090	1990	1895	1805	1725
	6000	2150	2050	1945	1845	1760	1680
	8000	2110	2000	1895	1795	1730	1625
6579 lbs (2984 kg)	SL	1820	1730	1650	1570	1490	1415
	2000	1780	1690	1600	1530	1460	1380
	4000	1740	1650	1560	1490	1410	1345
	6000	1700	1610	1520	1450	1370	1305
	8000	1660	1570	1480	1400	1330	1255

Figure 5.9.2 - CLIMB SPEEDS (IAS = 160 KIAS)

CLIMB PERFORMANCE

TIME, CONSUMPTION AND CLIMB DISTANCE (IAS = 130 KIAS)

Conditions : ***ISA - 20°C***
Maximum climb power
Landing gear and flaps UP
IAS = 130 KIAS - 2000 RPM - BLEED ON

NOTE :
Time, consumption and distance from the 50 ft

Pressure altitude (feet)	WEIGHT 4850 lbs (2200 kg)					WEIGHT 5512 lbs (2500 kg)					WEIGHT 6579 lbs (2984 kg)				
	Time (min. s)	Consump.			Dist. (NM)	Time (min. s)	Consump.			Dist. (NM)	Time (min. s)	Consump.			Dist. (NM)
		l	kg	us gal			l	kg	us gal			l	kg	us gal	
SL	00.00	0	0	0	0	00.00	0	0	0	0	00.00	0	0	0	0
2000	00.45	4	3	1.1	1	00.45	4	3	1.1	2	01.00	5	4	1.3	2
4000	01.30	6	5	1.6	3	01.30	8	6	2.1	3	02.00	10	8	2.6	4
6000	02.00	10	7	2.6	4	02.15	11	9	2.9	5	03.00	14	11	3.7	6
8000	02.45	12	10	3.2	6	03.00	15	12	4.0	7	04.00	19	15	5	9
10000	03.15	15	12	4.0	7	04.00	18	14	4.8	9	05.00	23	18	6	11
12000	04.00	18	14	4.8	9	04.45	21	17	5.5	11	06.00	27	22	7.1	14
14000	04.45	21	17	5.5	11	05.30	25	19	6.6	13	07.15	32	25	8.5	16
16000	05.30	24	19	6.3	13	06.30	28	22	7.4	15	08.15	36	28	9.5	19
18000	06.00	27	21	7.1	15	07.15	31	25	8.2	17	09.30	40	32	10.6	22
20000	07.00	29	23	7.7	17	08.00	35	27	9.2	20	10.30	44	35	11.6	25
22000	07.45	32	25	8.5	19	09.00	38	30	10.0	22	11.45	49	38	12.9	29
24000	08.30	35	27	9.3	21	10.00	41	32	10.8	25	13.00	53	42	14	32
26000	09.15	37	29	9.8	23	11.00	44	35	11.6	28	14.00	57	45	15.1	36
28000	10.00	40	32	10.6	26	12.00	48	37	12.7	31	15.30	62	49	16.4	40
30000	11.00	43	34	11.4	28	13.00	51	40	13.5	34	16.45	67	52	17.7	45
31000	11.24	45	35	12.0	31	13.34	54	42	14.0	37	17.48	70	55	19.0	48

Figure 5.9.3 - TIME, CONSUMPTION AND CLIMB DISTANCE
(IAS = 130 KIAS) / ISA - 20°C

CLIMB PERFORMANCE

TIME, CONSUMPTION AND CLIMB DISTANCE (IAS = 130 KIAS)

Conditions : ***ISA***
Maximum climb power
Landing gear and flaps UP
IAS = 130 KIAS - 2000 RPM - BLEED ON

NOTE :
Time, consumption and distance from the 50 ft

Pressure altitude (feet)	WEIGHT 4850 lbs (2200 kg)					WEIGHT 5512 lbs (2500 kg)					WEIGHT 6579 lbs (2984 kg)				
	Time (min. s)	Consump. l	Consump. kg	Consump. us gal	Dist. (NM)	Time (min. s)	Consump. l	Consump. kg	Consump. us gal	Dist. (NM)	Time (min. s)	Consump. l	Consump. kg	Consump. us gal	Dist. (NM)
SL	00.00	0	0	0	0	00.00	0	0	0	0	00.00	0	0	0	0
2000	00.45	4	3	1.1	2	00.45	4	3	1.1	2	01.00	5	4	1.3	2
4000	01.30	7	6	1.8	3	01.45	8	7	2.1	4	02.15	11	9	2.9	5
6000	02.15	11	8	2.9	5	02.30	12	10	3.2	6	03.15	16	13	4.2	7
8000	03.00	14	11	3.7	7	03.30	16	13	4.2	8	04.30	21	17	5.5	10
10000	03.45	17	14	4.5	8	04.15	20	16	5.3	10	05.30	26	20	6.9	13
12000	04.30	20	16	5.3	10	05.15	24	19	6.3	12	06.45	31	24	8.2	16
14000	05.00	24	19	6.3	12	06.15	28	22	7.4	15	08.00	36	28	9.5	19
16000	06.00	27	21	7.1	14	07.00	32	25	8.5	17	09.00	41	32	10.8	22
18000	06.45	30	23	7.9	17	08.00	35	28	9.2	20	10.30	46	36	12.2	26
20000	07.30	33	26	8.7	19	09.00	39	31	10.3	23	11.45	50	40	13.2	29
22000	08.30	36	28	9.5	21	10.00	43	34	11.4	26	13.00	55	43	14.5	33
24000	09.15	39	31	10.3	24	11.00	47	37	12.4	29	14.30	60	47	15.9	38
26000	10.15	43	33	11.4	27	12.15	51	40	13.5	33	16.00	66	52	17.4	43
28000	11.30	46	36	12.1	31	13.45	55	43	14.5	37	18.00	72	57	19	49
30000	12.45	50	39	13.2	36	15.30	60	47	15.9	43	20.30	79	62	20.9	58
31000	13.48	53	42	14.0	39	16.39	64	50	17.0	48	22.33	85	67	23.0	65

Figure 5.9.4 - TIME, CONSUMPTION AND CLIMB DISTANCE
(IAS = 130 KIAS) / ISA

CLIMB PERFORMANCE

TIME, CONSUMPTION AND CLIMB DISTANCE (IAS = 130 KIAS)

Conditions : ***ISA + 20°C***
Maximum climb power
Landing gear and flaps UP
IAS = 130 KIAS - 2000 RPM - BLEED ON

NOTE :
Time, consumption and distance from the 50 ft

Pressure altitude (feet)	WEIGHT 4850 lbs (2200 kg)					WEIGHT 5512 lbs (2500 kg)					WEIGHT 6579 lbs (2984 kg)				
	Time (min. s)	Consump.			Dist. (NM)	Time (min. s)	Consump.			Dist. (NM)	Time (min. s)	Consump.			Dist. (NM)
		l	kg	us gal			l	kg	us gal			l	kg	us gal	
SL	00.00	0	0	0	0	00.00	0	0	0	0	00.00	0	0	0	0
2000	00.45	4	3	1.0	2	01.00	5	4	1.3	2	01.15	6	5	1.6	3
4000	01.30	8	6	2.1	4	01.45	9	7	2.4	4	02.15	12	10	3.2	5
6000	02.15	12	9	3.2	5	02.45	14	11	3.7	6	03.30	18	14	4.8	8
8000	03.00	16	12	4.2	7	03.45	18	14	4.8	9	04.45	24	19	6.3	11
10000	04.00	19	15	5.0	9	04.45	23	18	6.1	11	06.00	29	23	7.7	15
12000	05.00	23	18	6.1	12	05.45	27	21	7.1	14	07.30	35	27	9.2	18
14000	05.45	26	21	6.9	14	06.45	31	24	8.2	17	08.45	40	32	10.6	22
16000	06.30	30	23	7.9	16	07.45	35	28	9.2	19	10.00	46	36	12.2	25
18000	07.30	34	26	9.0	19	08.45	40	31	10.6	23	11.30	52	40	13.7	30
20000	08.30	37	29	9.8	22	10.00	44	35	11.6	27	13.15	58	45	15.3	35
22000	09.45	41	32	10.8	26	11.30	49	39	12.9	31	15.15	64	50	16.9	41
24000	11.00	45	36	11.9	30	13.00	54	43	14.3	36	17.30	72	56	19	48
26000	12.30	50	39	13.2	35	15.00	60	47	15.9	43	20.30	80	63	21.1	58
28000	14.30	55	43	14.5	42	17.30	67	52	17.7	51	24.30	91	72	24	72
30000	17.00	62	48	16.4	51	21.00	75	59	19.8	63	30.30	107	84	28.3	94
31000	19.25	68	53	18.0	60	24.35	84	66	22.0	77	40.50	130	102	34.0	129

Figure 5.9.5 - TIME, CONSUMPTION AND CLIMB DISTANCE
(IAS = 130 KIAS) / ISA + 20°C

CLIMB PERFORMANCE

TIME, CONSUMPTION AND CLIMB DISTANCE (IAS = 160 KIAS)

Conditions : ***ISA - 20°C***
Maximum climb power
Landing gear and flaps UP
IAS = 160 KIAS up to 20000 ft ; - 2 KIAS / 1000 ft then
2000 RPM - BLEED ON

NOTE :
Time, consumption and distance from the 50 ft

Pressure altitude (feet)	WEIGHT 4850 lbs (2200 kg)					WEIGHT 5512 lbs (2500 kg)					WEIGHT 6579 lbs (2984 kg)				
	Time (min. s)	Consump. l	Consump. kg	Consump. us gal	Dist. (NM)	Time (min. s)	Consump. l	Consump. kg	Consump. us gal	Dist. (NM)	Time (min. s)	Consump. l	Consump. kg	Consump. us gal	Dist. (NM)
SL	00.00	0	0	0	0	00.00	0	0	0	0	00.00	0	0	0	0
2000	00.45	4	3	1.1	2	01.00	4	3	1.1	2	01.00	6	4	1.6	3
4000	01.30	7	5	1.8	4	01.45	9	7	2.4	5	02.15	11	9	2.9	6
6000	02.15	11	9	2.9	6	02.45	13	10	3.4	7	03.30	16	13	4.2	9
8000	03.00	14	11	3.7	8	03.30	17	13	4.5	10	04.30	21	17	5.5	12
10000	04.00	18	14	4.8	11	04.30	21	16	5.5	13	05.45	27	21	7.1	16
12000	04.45	21	17	5.5	13	05.30	25	20	6.6	15	07.00	32	25	8.5	20
14000	05.30	25	19	6.6	16	06.30	29	23	7.7	19	08.15	37	29	9.8	24
16000	06.30	28	22	7.4	18	07.30	33	26	8.7	22	09.30	42	33	11.1	28
18000	07.15	31	25	8.2	21	08.30	37	29	9.8	25	11.00	47	37	12.4	32
20000	08.15	35	27	9.2	24	09.45	41	32	10.8	29	12.30	52	41	13.7	37
22000	09.00	38	30	10.0	28	10.45	45	35	11.9	33	13.45	58	45	15.3	42
24000	10.00	41	32	10.8	31	12.00	49	39	12.9	37	15.15	63	49	16.6	47
26000	11.00	45	35	11.9	34	13.00	53	41	14.0	41	16.45	68	53	18	53
28000	12.00	48	37	12.7	37	14.00	56	44	14.8	44	18.15	73	57	19.3	58
30000	13.00	51	40	13.5	41	15.00	60	47	15.9	48	19.45	78	61	20.6	63
31000	13.27	53	42	14.0	43	15.59	63	50	17.0	51	20.52	82	64	22.0	67

Figure 5.9.6 - TIME, CONSUMPTION AND CLIMB DISTANCE
(IAS = 160 KIAS) / ISA - 20°C

CLIMB PERFORMANCE

TIME, CONSUMPTION AND CLIMB DISTANCE (IAS = 160 KIAS)

Conditions : ***ISA***
Maximum climb power
Landing gear and flaps UP
IAS = 160 KIAS up to 20000 ft ; - 2 KIAS / 1000 ft then
2000 RPM - BLEED ON

NOTE :
Time, consumption and distance from the 50 ft

Pressure altitude (feet)	WEIGHT 4850 lbs (2200 kg)					WEIGHT 5512 lbs (2500 kg)					WEIGHT 6579 lbs (2984 kg)				
	Time (min. s)	Consump.			Dist. (NM)	Time (min. s)	Consump.			Dist. (NM)	Time (min. s)	Consump.			Dist. (NM)
		l	kg	us gal			l	kg	us gal			l	kg	us gal	
SL	00.00	0	0	0	0	00.00	0	0	0	0	00.00	0	0	0	0
2000	00.45	4	3	1.1	2	01.00	5	4	1.3	3	01.15	6	5	1.6	3
4000	01.30	8	6	2.1	4	02.00	10	8	2.6	5	02.30	12	10	3.2	7
6000	02.30	12	10	3.2	7	03.00	14	11	3.7	8	03.45	18	15	4.8	10
8000	03.30	16	13	4.2	10	04.00	19	15	5.0	11	05.00	24	19	6.3	14
10000	04.30	20	16	5.3	12	05.00	24	19	6.3	14	06.30	30	24	7.9	18
12000	05.15	24	19	6.3	15	06.00	28	22	7.4	18	08.00	36	28	9.5	23
14000	06.15	28	22	7.4	18	07.15	33	26	8.7	21	09.15	42	33	11.1	28
16000	07.15	32	25	8.5	21	08.30	37	29	9.8	25	10.45	48	38	12.7	33
18000	08.15	36	28	9.5	25	09.45	42	33	11.1	29	12.30	54	42	14.3	38
20000	09.15	40	31	10.6	29	11.00	47	37	12.4	34	14.00	60	47	15.9	44
22000	10.15	44	34	11.6	32	12.00	52	41	13.7	39	15.45	67	52	17.7	50
24000	11.15	47	37	12.4	36	13.30	56	44	14.8	44	17.30	73	57	19.3	56
26000	12.30	51	40	13.5	41	15.00	62	49	16.4	50	19.30	79	62	20.9	64
28000	14.00	56	44	14.8	46	16.30	66	52	17.4	55	21.45	87	68	23.0	72
30000	15.30	60	47	15.8	52	18.30	72	57	19.0	62	24.30	95	75	25.1	83
31000	16.37	64	50	17.0	56	20.02	77	60	20.0	68	27.07	103	81	27.0	92

Figure 5.9.7 - TIME, CONSUMPTION AND CLIMB DISTANCE
(IAS = 160 KIAS) / ISA

CLIMB PERFORMANCE

TIME, CONSUMPTION AND CLIMB DISTANCE (IAS = 160 KIAS)

Conditions : ***ISA + 20°C***
Maximum climb power
Landing gear and flaps UP
IAS = 160 KIAS up to 20000 ft ; - 2 KIAS / 1000 ft then
2000 RPM - BLEED ON

NOTE :
Time, consumption and distance from the 50 ft

Pressure altitude (feet)	WEIGHT 4850 lbs (2200 kg)					WEIGHT 5512 lbs (2500 kg)					WEIGHT 6579 lbs (2984 kg)				
	Time (min. s)	Consump.			Dist. (NM)	Time (min. s)	Consump.			Dist. (NM)	Time (min. s)	Consump.			Dist. (NM)
		l	kg	us gal			l	kg	us gal			l	kg	us gal	
SL	00.00	0	0	0	0	00.00	0	0	0	0	00.00	0	0	0	0
2000	01.00	5	4	1.3	3	01.00	5	4	1.3	3	01.15	7	6	1.8	4
4000	01.45	9	7	2.4	5	02.15	11	8	2.9	6	02.45	14	11	3.7	8
6000	02.45	14	11	3.7	8	03.15	16	13	4.2	9	04.15	21	16	5.5	12
8000	03.45	18	14	4.8	11	04.30	21	17	5.5	13	05.30	28	22	7.4	16
10000	04.45	23	18	6.1	14	05.30	27	21	7.1	17	07.15	34	27	9.0	21
12000	05.45	27	21	7.1	17	06.45	32	25	8.5	21	08.45	41	32	10.8	26
14000	06.45	32	25	8.5	21	08.00	37	29	9.8	25	10.15	48	38	12.7	32
16000	08.00	36	28	9.5	25	09.30	43	33	11.4	29	12.00	55	43	14.5	38
18000	09.00	41	32	10.8	29	11.00	48	38	12.7	35	14.00	62	49	16.4	45
20000	10.30	46	36	12.2	34	12.30	55	43	14.5	41	16.15	71	56	18.8	53
22000	12.15	52	41	13.7	41	14.30	62	49	16.4	49	19.15	81	63	21.4	64
24000	14.15	58	45	15.3	48	17.00	69	54	18.2	58	22.45	92	72	24.3	78
26000	16.30	65	51	17.2	57	20.00	78	61	20.6	69	27.30	106	83	28.0	96
28000	19.15	72	56	19.0	68	23.45	88	69	23.2	84	34.00	124	98	32.8	123
30000	23.00	82	64	21.7	83	29.00	102	80	26.9	105	46.00	155	121	41.0	170
31000	26.06	90	70	24.0	94	34.05	115	90	30.0	124	69.31	212	166	56.0	259

Figure 5.9.8 - TIME, CONSUMPTION AND CLIMB DISTANCE
(IAS = 160 KIAS) / ISA + 20°C

CLIMB PERFORMANCE

CLIMB PERFORMANCE AFTER GO-AROUND

Conditions : Maximum climb power
Landing gear DN and flaps LDG
IAS = 90 KIAS

Airplane weight	Pressure altitude (feet)	RATE OF CLIMB (ft/min)						
		ISA - 35°C	ISA - 20°C	ISA - 10°C	ISA	ISA + 10°C	ISA + 20°C	ISA + 30°C
4850 lbs (2200 kg)	SL	2270	2100	2000	1910	1820	1740	1650
	2000	2240	2070	1960	1870	1780	1695	1620
	4000	2200	2030	1920	1830	1730	1650	1570
	6000	2160	1980	1880	1780	1690	1600	1520
	8000	2120	1940	1830	1730	1630	1545	1465
5512 lbs (2500 kg)	SL	1900	1750	1660	1580	1500	1435	1355
	2000	1860	1720	1630	1550	1470	1395	1330
	4000	1820	1680	1590	1500	1430	1350	1285
	6000	1790	1630	1550	1460	1380	1305	1235
	8000	1750	1590	1500	1410	1330	1255	1185
6579 lbs (2984 kg)	SL	1410	1300	1230	1165	1105	1045	985
	2000	1380	1265	1195	1130	1065	1010	955
	4000	1345	1230	1155	1090	1025	970	915
	6000	1310	1190	1115	1050	985	925	870
	8000	1270	1145	1070	1000	940	880	825

Figure 5.9.9 - CLIMB PERFORMANCE AFTER GO-AROUND

CLIMB PERFORMANCE

CLIMB PERFORMANCE - FLAPS TO

Conditions : Climb maximum power
Landing gear UP and flaps TO
IAS = 110 KIAS

Airplane weight	Pressure altitude (feet)	RATE OF CLIMB (ft/min)						
		ISA - 35°C	ISA - 20°C	ISA - 10°C	ISA	ISA + 10°C	ISA + 20°C	ISA + 30°C
4850 lbs (2200 kg)	SL	3170	2970	2850	2730	2620	2520	2410
	2000	3160	2950	2820	2700	2600	2490	2395
	4000	3140	2920	2800	2670	2570	2460	2365
	6000	3110	2900	2760	2650	2540	2430	2330
	8000	3080	2870	2740	2610	2500	2395	2295
5512 lbs (2500 kg)	SL	2710	2540	2430	2330	2230	2145	2050
	2000	2700	2520	2400	2300	2200	2120	2035
	4000	2680	2490	2380	2270	2180	2090	2005
	6000	2650	2460	2350	2250	2150	2060	1975
	8000	2620	2440	2320	2220	2120	2030	1940
6579 lbs (2984 kg)	SL	2140	2000	1910	1830	1750	1680	1600
	2000	2120	1975	1880	1800	1720	1650	1585
	4000	2100	1950	1860	1775	1700	1620	1555
	6000	2075	1925	1830	1750	1670	1595	1525
	8000	2050	1895	1805	1720	1640	1565	1495

Figure 5.9.10 - CLIMB PERFORMANCE - FLAPS TO

5.10 - CRUISE PERFORMANCE

Conditions : ISA
Weight 5512 lbs (2500 kg)

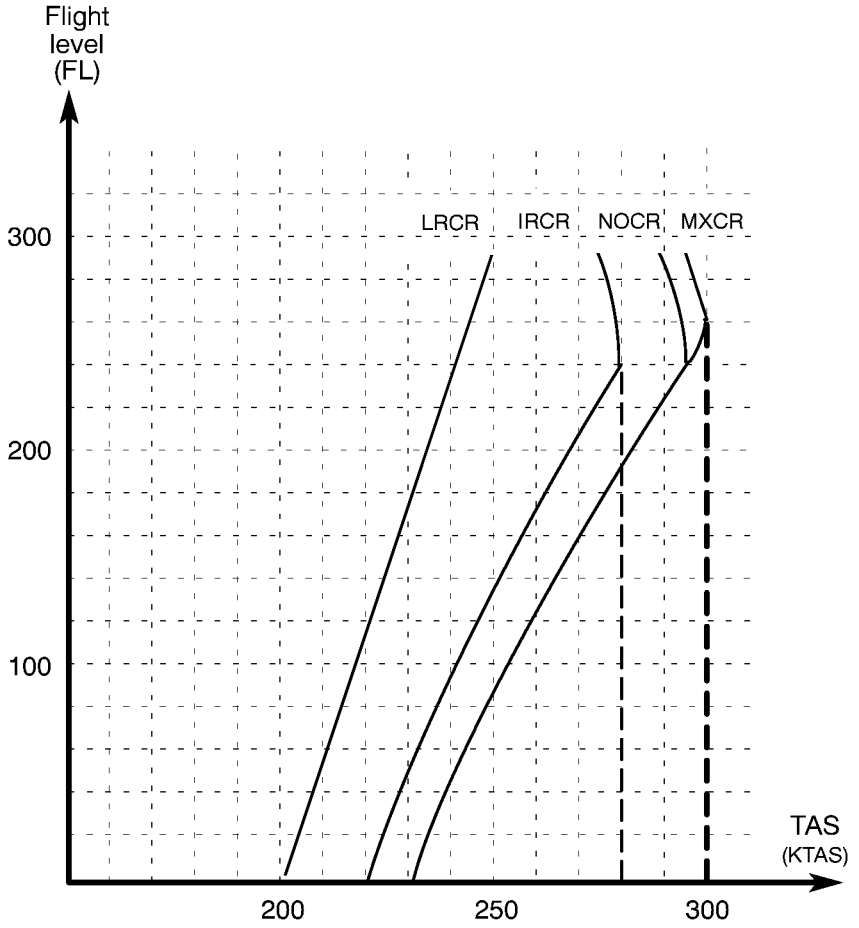


Figure 5.10.1 - CRUISE PERFORMANCE

CRUISE PERFORMANCE

Maximum cruise

Conditions : ***ISA - 20°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Use preferably recommended cruise power

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 2	100	304	239	80.3	231	226	230	225	229	224
5000	- 8	100	275	216	72.6	226	237	225	236	223	235
10000	- 17	100	250	196	66.0	221	249	220	248	218	246
15000	- 26	100	232	182	61.2	216	263	214	261	213	259
18000	- 32	100	223	175	58.9	213	271	211	269	210	267
20000	- 36	100	218	171	57.6	211	277	209	275	208	273
21000	- 37	100	216	170	57.1	210	280	208	278	207	276
22000	- 39	100	214	168	56.5	209	283	207	281	206	279
23000	- 41	100	212	166	56.0	208	286	206	284	205	282
24000	- 43	100	210	165	55.6	207	290	205	287	203	285
25000	- 45	100	209	164	55.3	206	293	204	291	202	288
26000	- 46	100	208	163	54.9	205	296	203	294	201	291
27000	- 48	100	207	162	54.7	204	300	202	297	200	294
28000	- 50	100	206	162	54.4	203	303	201	301	199	298
29000	- 52	100	206	161	54.3	202	307	200	304	198	301
30000	- 54	100	205	161	54.2	201	310	199	308	197	305
31000	- 56	100	206	161	54.3	200	314	198	311	196	308

Figure 5.10.2 - CRUISE PERFORMANCE -
Maximum cruise / ISA - 20°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Maximum cruise

Conditions : ***ISA - 10°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Use preferably recommended cruise power

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 12	100	308	242	81.4	229	228	228	227	227	226
5000	+ 2	100	279	219	73.6	224	240	223	238	222	237
10000	- 7	100	254	199	67.0	219	252	218	251	216	249
15000	- 16	100	234	184	61.9	214	266	213	264	211	262
18000	- 22	100	225	177	59.4	211	274	209	272	208	270
20000	- 25	100	220	173	58.2	209	281	207	278	206	276
21000	- 27	100	218	171	57.6	208	284	206	281	205	279
22000	- 29	100	216	170	57.1	207	287	205	285	204	282
23000	- 31	100	215	168	56.7	206	290	204	288	202	285
24000	- 33	100	213	167	56.3	205	293	203	291	201	288
25000	- 34	100	212	166	55.9	204	297	202	294	200	291
26000	- 36	100	210	165	55.6	203	300	201	298	199	295
27000	- 38	100	209	164	55.3	202	304	200	301	198	298
28000	- 40	100	209	164	55.2	201	307	199	305	197	302
29000	- 42	97	201	158	53.2	197	307	195	303	193	300
30000	- 44	93	194	152	51.2	193	306	190	301	188	298
31000	- 46	91	187	146	49.3	189	304	186	300	184	296

Figure 5.10.3 - CRUISE PERFORMANCE -
Maximum cruise / ISA - 10°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Maximum cruise

Conditions : ***ISA - 5°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Use preferably recommended cruise power

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 17	100	310	243	81.8	228	230	227	229	226	228
5000	+ 8	100	280	220	74.1	223	241	222	240	221	238
10000	- 2	100	255	200	67.4	218	254	217	252	216	250
15000	- 11	100	235	185	62.2	213	267	212	265	210	263
18000	- 17	100	226	178	59.8	210	276	208	274	207	272
20000	- 20	100	222	174	58.6	208	282	206	280	205	278
21000	- 22	100	220	173	58.1	207	285	205	283	204	281
22000	- 24	100	218	171	57.5	206	289	204	286	203	284
23000	- 26	100	216	170	57.1	205	292	203	289	202	287
24000	- 28	100	215	168	56.7	204	295	202	293	200	290
25000	- 29	100	213	167	56.3	203	298	201	296	199	293
26000	- 31	100	212	166	56.0	202	302	200	299	198	296
27000	- 33	99	210	165	55.5	200	304	198	301	196	299
28000	- 35	96	202	159	53.5	197	304	194	301	193	298
29000	- 37	92	195	153	51.5	193	303	190	300	188	296
30000	- 39	88	188	147	49.6	188	302	186	298	183	294
31000	- 41	86	181	142	47.7	184	300	182	296	179	292

Figure 5.10.4 - CRUISE PERFORMANCE -
Maximum cruise / ISA - 5°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Maximum cruise

Conditions : ***ISA***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Use preferably recommended cruise power

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 22	100	312	245	82.3	228	231	227	230	226	229
5000	+ 13	100	282	221	74.5	223	243	221	241	220	240
10000	+ 3	100	257	201	67.8	217	255	216	253	215	252
15000	- 6	100	237	186	62.5	212	269	211	267	209	265
18000	- 12	100	228	179	60.2	209	278	208	275	206	273
20000	- 15	100	223	175	58.9	207	284	205	281	204	279
21000	- 17	100	221	174	58.5	206	287	204	285	203	282
22000	- 19	100	220	172	58.0	205	290	203	288	202	285
23000	- 21	100	218	171	57.5	204	293	202	291	201	289
24000	- 22	100	216	170	57.1	203	297	201	294	199	292
25000	- 24	100	215	169	56.8	202	300	200	298	198	295
26000	- 26	99	209	164	55.2	200	303	198	300	197	298
27000	- 28	95	202	159	53.5	196	302	194	298	193	297
28000	- 30	91	195	153	51.6	192	301	189	297	188	295
29000	- 32	88	188	148	49.8	188	299	186	296	184	293
30000	- 34	84	181	142	47.9	184	298	182	294	178	288
31000	- 36	81	174	136	45.9	180	296	177	292	174	287

Figure 5.10.5 - CRUISE PERFORMANCE -
Maximum cruise / ISA

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Maximum cruise

Conditions : ***ISA + 5°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Use preferably recommended cruise power

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 27	100	313	246	82.8	227	232	226	231	225	230
5000	+ 18	100	283	223	74.9	222	244	221	243	219	241
10000	+ 8	100	258	202	68.1	217	256	215	255	214	253
15000	- 1	100	238	187	62.9	211	270	210	268	208	267
18000	- 6	100	229	180	60.6	208	279	207	277	205	275
20000	- 10	100	224	176	59.3	206	285	204	283	203	281
21000	- 12	100	223	175	58.9	205	288	203	286	202	284
22000	- 14	100	221	174	58.5	204	291	202	289	201	287
23000	- 16	100	220	172	58.0	203	295	201	292	200	290
24000	- 17	100	218	172	57.7	202	298	200	296	198	293
25000	- 19	97	211	166	55.9	199	299	197	296	195	294
26000	- 21	94	204	161	54.0	196	299	193	296	192	293
27000	- 23	90	197	155	52.0	192	298	189	294	188	292
28000	- 25	87	190	149	50.1	188	297	185	293	183	290
29000	- 27	83	182	143	48.2	183	296	181	292	178	287
30000	- 29	80	176	138	46.4	179	294	177	290	173	284
31000	- 31	76	168	132	44.5	175	293	172	288	169	282

Figure 5.10.6 - CRUISE PERFORMANCE -
Maximum cruise / ISA + 5°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Maximum cruise

Conditions : ***ISA + 10°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Use preferably recommended cruise power

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 32	100	315	247	83.2	226	233	225	233	224	231
5000	+ 23	100	285	224	75.4	221	245	220	244	219	242
10000	+ 13	100	259	203	68.4	216	258	214	256	213	255
15000	+ 4	100	240	188	63.3	211	272	209	270	208	268
18000	- 1	100	231	181	60.9	208	281	206	278	204	277
20000	- 5	100	226	177	59.7	206	287	203	284	202	283
21000	- 7	100	224	176	59.2	205	291	202	287	201	286
22000	- 9	100	223	175	58.9	203	294	201	291	200	289
23000	- 11	98	218	171	57.6	201	295	200	294	197	290
24000	- 13	96	211	166	55.7	198	296	196	293	194	290
25000	- 15	92	204	160	53.9	195	296	192	292	191	290
26000	- 17	89	197	155	52.0	191	296	188	291	186	288
27000	- 19	86	190	150	50.3	187	295	185	290	182	287
28000	- 20	82	184	144	48.5	184	294	181	289	178	285
29000	- 22	79	176	139	46.6	179	292	176	287	173	282
30000	- 24	75	170	133	44.9	175	290	172	285	168	279
31000	- 26	72	163	128	43.0	171	288	167	283	163	276

Figure 5.10.7 - CRUISE PERFORMANCE -
Maximum cruise / ISA + 10°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Maximum cruise

Conditions : ***ISA + 20°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Use preferably recommended cruise power

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 42	100	319	250	84.3	225	236	224	235	223	233
5000	+ 33	100	289	227	76.3	220	248	218	246	217	245
10000	+ 23	100	262	206	69.3	214	261	213	259	211	257
15000	+ 14	100	243	190	64.1	209	275	207	273	206	270
18000	+ 9	100	234	183	61.7	206	284	204	281	202	279
20000	+ 4	97	225	177	59.4	202	288	201	287	199	284
21000	+ 2	94	218	171	57.5	198	288	198	286	196	283
22000	0	92	211	166	55.7	196	288	193	285	191	282
23000	- 2	88	204	160	53.9	195	292	190	284	188	281
24000	- 3	86	197	155	52.0	188	287	185	283	184	280
25000	- 5	82	190	149	50.2	185	286	182	282	179	278
26000	- 7	79	183	144	48.3	181	285	178	281	175	276
27000	- 9	76	176	139	46.6	176	283	174	279	170	274
28000	- 11	72	170	133	44.8	172	281	169	276	166	272
29000	- 13	69	163	128	43.1	168	280	164	274	161	268
30000	- 15	66	156	122	41.2	164	278	160	272	156	265
31000	- 17	62	150	117	39.5	159	275	155	268	150	260

Figure 5.10.8 - CRUISE PERFORMANCE -
Maximum cruise / ISA + 20°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Normal (recommended) cruise

Conditions : ***ISA - 20°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Power recommended by PRATT & WHITNEY CANADA

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 2	100	304	239	80.3	231	226	230	225	229	224
5000	- 8	100	275	216	72.6	226	237	225	236	223	235
10000	- 17	100	250	196	66.0	221	249	220	248	218	246
15000	- 26	100	232	182	61.2	216	263	214	261	213	259
18000	- 32	100	223	175	58.9	213	271	211	269	210	267
20000	- 36	100	218	171	57.6	211	277	209	275	208	273
21000	- 37	100	216	170	57.1	210	280	208	278	207	276
22000	- 39	100	214	168	56.5	209	283	207	281	206	279
23000	- 41	100	212	166	56.0	208	286	206	284	205	282
24000	- 43	100	210	165	55.6	207	290	205	287	203	285
25000	- 45	100	209	164	55.3	206	293	204	291	202	288
26000	- 46	100	208	163	54.9	205	296	203	294	201	291
27000	- 48	100	207	162	54.7	204	300	202	297	200	294
28000	- 50	100	206	162	54.4	203	303	201	301	199	298
29000	- 52	100	206	161	54.3	202	307	200	304	198	301
30000	- 54	100	205	161	54.2	201	310	199	308	197	305
31000	- 56	95	198	155	52.2	197	310	195	306	193	303

Figure 5.10.9 - CRUISE PERFORMANCE -
Normal cruise / ISA - 20°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Normal (recommended) cruise

Conditions : ***ISA - 10°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Power recommended by PRATT & WHITNEY CANADA

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 12	100	308	242	81.4	229	228	228	227	227	226
5000	+ 2	100	279	219	73.6	224	240	223	238	222	237
10000	- 7	100	254	199	67.0	219	252	218	251	216	249
15000	- 16	100	234	184	61.9	214	266	213	264	211	262
18000	- 22	100	225	177	59.4	211	274	209	272	208	270
20000	- 25	100	220	173	58.2	209	281	207	278	206	276
21000	- 27	100	218	171	57.6	208	284	206	281	205	279
22000	- 29	100	216	170	57.1	207	287	205	285	204	282
23000	- 31	100	215	168	56.7	206	290	204	288	202	285
24000	- 33	100	213	167	56.3	205	293	203	291	201	288
25000	- 34	100	212	166	55.9	204	297	202	294	200	291
26000	- 36	100	210	165	55.6	203	300	201	298	199	295
27000	- 38	99	207	162	54.7	201	302	199	299	196	296
28000	- 40	96	199	157	52.7	197	302	195	298	192	294
29000	- 42	92	193	151	50.9	193	300	191	297	188	293
30000	- 44	88	185	145	48.9	188	298	186	295	184	291
31000	- 46	85	179	140	47.2	185	298	182	294	179	289

Figure 5.10.10 - CRUISE PERFORMANCE -
Normal cruise / ISA - 10°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Normal (recommended) cruise

Conditions : ***ISA - 5°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Power recommended by PRATT & WHITNEY CANADA

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 17	100	310	243	81.8	228	230	227	229	226	228
5000	+ 8	100	280	220	74.1	223	241	222	240	221	238
10000	- 2	100	255	200	67.4	218	254	217	252	216	250
15000	- 11	100	235	185	62.2	213	267	212	265	210	263
18000	- 17	100	226	178	59.8	210	276	208	274	207	272
20000	- 20	100	222	174	58.6	208	282	206	280	205	278
21000	- 22	100	220	173	58.1	207	285	205	283	204	281
22000	- 24	100	218	171	57.5	206	289	204	286	203	284
23000	- 26	100	216	170	57.1	205	292	203	289	202	287
24000	- 28	100	215	168	56.7	204	295	202	293	200	290
25000	- 29	100	213	167	56.3	203	298	201	296	199	293
26000	- 31	98	208	163	54.9	200	299	198	295	196	293
27000	- 33	95	201	158	53.1	196	298	194	296	192	292
28000	- 35	91	195	153	51.4	193	298	190	295	187	290
29000	- 37	87	188	147	49.6	188	297	186	293	183	288
30000	- 39	83	181	142	47.7	184	295	182	291	179	287
31000	- 41	80	172	135	45.4	180	293	177	289	174	284

Figure 5.10.11 - CRUISE PERFORMANCE -
Normal cruise / ISA - 5°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Normal (recommended) cruise

Conditions : ***ISA***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Power recommended by PRATT & WHITNEY CANADA

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 22	100	312	245	82.3	228	231	227	230	226	229
5000	+ 13	100	282	221	74.5	223	243	221	241	220	240
10000	+ 3	100	257	201	67.8	217	255	216	253	215	252
15000	- 6	100	237	186	62.5	212	269	211	267	209	265
18000	- 12	100	228	179	60.2	209	278	208	275	206	273
20000	- 15	100	223	175	58.9	207	284	205	281	204	279
21000	- 17	100	221	174	58.5	206	287	204	285	203	282
22000	- 19	100	220	172	58.0	205	290	203	288	202	285
23000	- 21	100	218	171	57.5	204	293	202	291	201	289
24000	- 22	100	216	170	57.1	203	297	201	294	199	292
25000	- 24	97	209	164	55.3	199	296	197	294	195	291
26000	- 26	94	203	159	53.6	195	295	194	293	191	290
27000	- 28	90	196	154	51.9	192	295	190	292	187	287
28000	- 31	86	190	149	50.1	188	294	186	291	183	286
29000	- 33	83	183	144	48.3	184	293	182	289	178	284
30000	- 35	78	176	138	46.5	180	291	177	287	174	282
31000	- 37	76	166	130	43.9	176	290	173	285	169	279

Figure 5.10.12 - CRUISE PERFORMANCE -
Normal cruise / ISA

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Normal (recommended) cruise

Conditions : ***ISA + 5°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Power recommended by PRATT & WHITNEY CANADA

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 27	100	313	246	82.8	227	232	226	231	225	230
5000	+ 18	100	283	223	74.9	222	244	221	243	219	241
10000	+ 8	100	258	202	68.1	217	256	215	255	214	253
15000	- 1	100	238	187	62.9	211	270	210	268	208	267
18000	- 6	100	229	180	60.6	208	279	207	277	205	275
20000	- 10	100	224	176	59.3	206	286	204	283	203	281
21000	- 12	100	223	175	58.9	205	288	203	286	202	284
22000	- 14	100	221	174	58.5	204	291	202	289	201	287
23000	- 16	97	216	170	57.1	201	292	199	290	198	287
24000	- 18	95	209	164	55.1	198	292	196	290	194	287
25000	- 20	92	202	159	53.4	194	292	192	289	190	286
26000	- 22	89	195	153	51.5	190	291	189	289	186	285
27000	- 24	84	188	148	49.8	187	290	185	287	181	283
28000	- 26	81	182	143	48.0	183	290	180	286	177	281
29000	- 28	78	175	137	46.2	179	288	176	284	172	278
30000	- 30	74	171	134	45.2	174	286	171	281	168	276
31000	- 32	71	161	126	42.5	171	285	168	280	164	274

Figure 5.10.13 - CRUISE PERFORMANCE -
Normal cruise / ISA + 5°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Normal (recommended) cruise

Conditions : ***ISA + 10°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Power recommended by PRATT & WHITNEY CANADA

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 32	100	315	247	83.2	226	233	225	233	224	231
5000	+ 23	100	285	224	75.4	221	245	220	244	219	242
10000	+ 13	100	259	203	68.4	216	258	214	256	213	255
15000	+ 4	100	240	188	63.3	211	272	209	270	208	268
18000	- 1	100	231	181	60.9	208	281	206	278	204	277
20000	- 6	100	226	177	59.7	206	287	203	284	202	283
21000	- 8	98	220	173	58.1	203	288	202	286	199	283
22000	- 10	96	214	168	56.5	200	289	198	286	196	284
23000	- 12	92	207	162	54.7	197	289	195	286	192	283
24000	- 13	90	200	157	52.8	193	289	191	286	188	282
25000	- 15	87	193	152	51.1	190	288	187	285	185	281
26000	- 17	83	187	147	49.4	185	287	184	284	181	279
27000	- 19	79	181	142	47.7	182	286	179	282	176	277
28000	- 21	76	174	137	46.0	178	285	175	280	172	275
29000	- 23	73	167	131	44.1	173	283	170	278	167	272
30000	- 25	69	166	130	43.9	169	280	166	276	162	269
31000	- 27	67	154	121	40.7	165	279	162	273	157	266

Figure 5.10.14 - CRUISE PERFORMANCE -
Normal cruise / ISA + 10°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Normal (recommended) cruise

Conditions : ***ISA + 20°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

NOTE :
Power recommended by PRATT & WHITNEY CANADA

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 42	100	319	250	84.3	225	236	224	235	223	233
5000	+ 33	100	289	227	76.3	219	248	218	246	217	245
10000	+ 23	100	262	206	69.3	214	261	213	259	211	257
15000	+ 14	100	243	190	64.1	209	275	207	273	206	270
18000	+ 9	96	226	177	59.7	202	279	201	278	199	275
20000	+ 4	90	213	167	56.3	196	280	194	277	193	275
21000	+ 2	87	206	162	54.4	193	279	191	277	189	274
22000	0	84	200	157	52.8	189	279	187	276	185	274
23000	- 2	81	193	152	51.0	185	278	184	275	182	272
24000	- 4	78	187	146	49.3	181	276	180	274	178	271
25000	- 6	76	180	141	47.6	177	275	176	273	173	269
26000	- 8	72	173	136	45.8	173	273	172	271	169	267
27000	- 10	69	167	131	44.1	169	272	168	270	164	264
28000	- 12	66	160	126	42.3	164	269	163	268	159	261
29000	- 14	63	157	120	40.6	160	268	159	265	154	257
30000	- 16	60	147	115	38.8	156	265	154	262	149	254
31000	- 18	58	141	111	37.3	153	265	148	257	/	/

Figure 5.10.15 - CRUISE PERFORMANCE -
Normal cruise / ISA + 20°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Intermediate cruise

Conditions : ***ISA - 20°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 1	88	286	225	75.6	220	215	219	214	218	213
5000	- 8	88	257	202	68.0	215	226	213	222	212	221
10000	- 18	88	233	183	61.6	210	238	209	236	208	234
15000	- 27	88	214	168	56.5	206	250	204	248	202	246
18000	- 32	88	205	161	54.2	203	258	201	256	199	254
20000	- 36	88	200	157	52.8	201	264	199	262	197	259
21000	- 38	88	198	155	52.2	200	267	198	265	196	262
22000	- 40	88	195	153	51.6	199	270	197	268	195	265
23000	- 42	88	193	152	51.1	198	273	196	271	194	268
24000	- 44	88	192	151	50.7	197	276	195	274	193	271
25000	- 45	88	190	149	50.2	196	279	194	277	192	274
26000	- 47	88	188	148	49.7	195	283	193	280	191	277
27000	- 49	88	187	147	49.4	194	286	192	283	190	280
28000	- 51	88	186	146	49.1	193	289	191	286	189	283
29000	- 53	88	185	145	48.9	192	293	190	290	188	286
30000	- 54	87	183	144	48.3	190	294	188	292	185	287
31000	- 56	84.5	178	139	46.9	187	294	185	291	182	287

Figure 5.10.16 - CRUISE PERFORMANCE -
Intermediate cruise / ISA - 20°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Intermediate cruise

Conditions : ***ISA - 10°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 11	88	289	227	76.3	219	218	218	217	217	216
5000	+ 2	88	261	205	68.9	214	229	212	226	210	224
10000	- 7	88	236	185	62.3	209	241	208	239	206	237
15000	- 17	88	217	170	57.2	204	254	203	252	201	250
18000	- 22	88	207	162	54.7	201	262	200	260	198	257
20000	- 26	88	202	159	53.4	199	268	198	265	196	263
21000	- 28	88	199	157	52.7	198	271	196	268	195	266
22000	- 30	88	198	155	52.2	197	274	196	271	193	268
23000	- 31	88	195	153	51.6	196	277	195	274	192	271
24000	- 33	88	194	152	51.2	195	280	193	277	191	274
25000	- 35	88	192	151	50.7	194	283	192	280	190	277
26000	- 37	88	190	150	50.3	192	285	191	283	188	279
27000	- 39	85	185	146	49.0	189	285	188	283	185	279
28000	- 41	82	179	141	47.3	186	285	184	281	182	278
29000	- 43	79	173	135	45.6	183	285	180	280	177	276
30000	- 45	76	166	130	43.9	179	283	176	279	173	274
31000	- 47	74	162	127	42.7	175	283	173	279	169	274

Figure 5.10.17 - CRUISE PERFORMANCE -
Intermediate cruise / ISA - 10°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Intermediate cruise

Conditions : ***ISA - 5°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 17	88	290	228	76.7	218	219	217	218	216	217
5000	+ 7	88	262	206	69.3	213	230	211	227	210	225
10000	- 2	88	237	186	62.7	208	242	207	241	205	239
15000	- 12	88	218	171	57.5	203	255	202	253	200	251
18000	- 17	88	209	164	55.1	200	263	199	261	197	259
20000	- 21	88	203	160	53.7	198	269	197	267	195	264
21000	- 23	88	201	158	53.1	197	272	196	270	194	267
22000	- 25	88	199	156	52.5	196	275	195	273	193	270
23000	- 26	88	197	155	52.0	195	278	194	276	191	273
24000	- 28	88	195	153	51.5	194	281	193	279	190	276
25000	- 30	87.5	193	151	50.9	192	283	191	281	188	278
26000	- 32	84.7	187	146	49.3	189	283	187	281	185	277
27000	- 34	82	181	142	47.7	185	283	184	280	181	276
28000	- 36	79	174	137	46.0	182	282	180	279	177	274
29000	- 38	76	167	131	44.2	178	281	175	277	172	272
30000	- 40	73.3	161	126	42.5	174	280	171	275	168	270
31000	- 42	69.5	156	122	41.1	170	278	167	273	163	267

Figure 5.10.18 - CRUISE PERFORMANCE -
Intermediate cruise / ISA - 5°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Intermediate cruise

Conditions : ***ISA***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 22	88	292	229	77.1	218	221	216	219	215	218
5000	+ 12	88	264	207	69.8	213	232	210	228	209	227
10000	+ 3	88	239	188	63.1	208	243	206	242	205	240
15000	- 7	88	219	172	57.9	203	257	201	255	199	253
18000	- 12	88	210	165	55.5	199	265	198	263	196	260
20000	- 16	88	204	161	54.0	197	271	196	269	194	266
21000	- 18	88	202	159	53.5	196	273	195	272	193	269
22000	- 20	88	200	157	52.8	195	277	194	275	192	272
23000	- 21	88	198	156	52.4	194	280	193	278	191	275
24000	- 23	87.3	195	153	51.5	193	283	191	280	188	276
25000	- 25	84.3	188	148	49.8	189	282	188	279	185	275
26000	- 27	81.2	182	143	48.1	185	281	184	278	181	274
27000	- 29	78	176	138	46.4	182	280	180	277	177	272
28000	- 31	75	169	133	44.6	178	279	176	276	172	270
29000	- 33	72	162	128	42.9	174	277	171	273	168	268
30000	- 35	69	156	122	41.2	170	276	167	271	163	265
31000	- 37	66	151	118	39.8	165	273	162	268	158	262

Figure 5.10.19 - CRUISE PERFORMANCE -
Intermediate cruise / ISA

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Intermediate cruise

Conditions : ***ISA + 5°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 27	88	293	230	77.5	217	222	216	221	215	220
5000	+ 17	88	266	209	70.2	212	233	209	229	208	228
10000	+ 8	88	240	188	63.4	207	245	205	243	204	242
15000	- 2	88	220	173	58.2	202	258	200	256	199	254
18000	- 7	88	211	166	55.8	199	266	197	264	195	262
20000	- 11	88	206	162	54.4	196	272	195	270	193	268
21000	- 13	88	204	160	53.9	195	275	194	273	192	270
22000	- 14	88	201	158	53.1	194	278	193	276	190	272
23000	- 16	85.6	195	153	51.5	191	279	189	276	187	272
24000	- 18	82.7	189	148	49.9	188	279	186	276	183	272
25000	- 20	79.8	182	143	48.2	185	278	182	275	180	271
26000	- 22	76.9	176	139	46.6	181	277	179	274	175	269
27000	- 24	74	170	134	45.0	177	275	174	272	172	269
28000	- 27	71	164	129	43.3	173	274	170	270	166	264
29000	- 29	68.1	157	124	41.6	169	273	166	268	162	262
30000	- 31	65.2	151	119	39.9	165	271	161	266	158	260
31000	- 33	62.5	146	114	38.5	160	269	157	263	152	256

Figure 5.10.20 - CRUISE PERFORMANCE -
Intermediate cruise / ISA + 5°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Intermediate cruise

Conditions : ***ISA + 10°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 32	88	295	232	77.9	216	223	215	222	214	221
5000	+ 22	88	267	210	70.6	211	234	209	230	208	229
10000	+ 13	88	241	189	63.7	206	246	205	245	203	243
15000	+ 4	88	221	174	58.5	201	259	199	258	198	255
18000	- 2	88	212	167	56.1	198	268	196	266	194	263
20000	- 6	88	207	162	54.6	196	274	194	272	192	269
21000	- 8	86.5	201	158	53.1	194	275	191	272	189	269
22000	- 10	84	195	153	51.5	190	275	188	272	186	269
23000	- 12	81	190	149	50.1	187	275	185	271	182	268
24000	- 14	78	183	144	48.3	184	275	181	271	178	267
25000	- 16	75.5	177	139	46.8	180	273	177	270	174	266
26000	- 18	73	171	134	45.2	176	272	174	269	170	263
27000	- 20	70	165	130	43.6	172	271	169	267	168	265
28000	- 22	67	159	124	41.9	168	269	165	265	161	258
29000	- 24	64	153	120	40.3	164	268	160	262	157	256
30000	- 26	61	146	115	38.6	160	266	156	260	153	255
31000	- 28	59	141	111	37.3	155	264	152	257	/	/

Figure 5.10.21 - CRUISE PERFORMANCE -
Intermediate cruise / ISA + 10°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Intermediate cruise

Conditions : ***ISA + 20°C***
Landing gear and flaps UP
2000 RPM (*) - BLEED ON

Pressure altitude (feet)	IOAT (°C)	TRQ (%)	Fuel flow			AIRSPEEDS (kt)					
						4850 lbs (2200 kg)		5512 lbs (2500 kg)		6173 lbs (2800 kg)	
			l / h	kg / h	us gal / h	IAS	TAS	IAS	TAS	IAS	TAS
0	+ 42	88	299	235	79.0	215	225	214	224	213	223
5000	+ 32	88	271	212	71.5	210	237	208	233	206	232
10000	+ 23	88	245	192	64.6	205	249	203	247	202	245
15000	+ 14	88	224	176	59.3	199	262	198	260	196	258
18000	+ 8	84	209	164	55.2	192	266	190	263	189	261
20000	+ 4	79	196	154	51.9	186	266	185	263	182	260
21000	+ 2	76.4	190	150	50.3	183	266	181	263	179	259
22000	0	74	184	144	48.6	180	265	178	262	175	258
23000	- 2	71.2	178	140	47.0	176	264	174	262	171	257
24000	- 4	69	172	135	45.4	173	264	171	261	167	255
25000	- 6	66	165	130	43.7	169	262	166	259	162	253
26000	- 8	63.4	159	125	42.1	165	261	162	256	158	250
27000	- 10	60.7	154	120	40.6	160	258	157	254	153	247
28000	- 12	58	148	116	39.0	157	257	153	251	148	243
29000	- 14	55.5	141	111	37.2	152	254	148	248	142	238
30000	- 17	53	135	106	35.7	148	252	143	244	136	232
31000	- 19	51	130	102	34.3	143	249	/	/	/	/

Figure 5.10.22 - CRUISE PERFORMANCE -
Intermediate cruise / ISA + 20°C

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Long Range Cruise (5512 lbs - 2500 kg)

Conditions : Landing gear and flaps UP
2000 RPM (*) - BLEED ON

LEGEND : **IOAT**: °C **IAS** : KIAS
FF : us gal/h
FF : lbs/h **TAS**: KTAS

Presssure altitude (feet)	TRQ (%)	ISA - 20°C	ISA - 10°C	ISA	ISA + 10°C	ISA + 20°C
15000	60.5	**- 28** **176** 46.4 304 **214**	**- 18** **174** 47.0 309 **216**	**- 8** **172** 47.6 311 **219**	**+ 2** **171** 48.2 315 **221**	**+ 12** **169** 48.9 320 **223**
18000	59.0	**- 34** **171** 43.2 282 **218**	**- 24** **169** 43.7 287 **221**	**- 14** **168** 44.4 291 **223**	**- 4** **166** 45.0 295 **226**	**+ 6** **165** 45.7 300 **228**
19000	58.5	**- 36** **169** 42.2 276 **220**	**- 26** **167** 42.7 280 **222**	**- 16** **166** 43.4 284 **225**	**- 6** **164** 44.1 289 **227**	**+ 4** **163** 44.7 293 **229**
20000	58.0	**- 38** **168** 41.2 269 **222**	**- 28** **166** 41.7 273 **224**	**- 18** **164** 42.4 278 **226**	**- 8** **163** 43.1 282 **228**	**+ 2** **161** 43.7 287 **231**
21000	57.5	**- 40** **166** 40.4 265 **223**	**- 30** **164** 40.8 267 **225**	**- 20** **163** 41.5 271 **228**	**- 10** **161** 42.1 276 **230**	**+ 0** **160** 42.7 280 **232**
22000	57.0	**- 42** **165** 39.5 258 **224**	**- 32** **163** 39.9 262 **227**	**- 22** **161** 40.6 265 **229**	**- 12** **159** 41.2 269 **232**	**- 1** **158** 41.7 273 **234**
23000	56.5	**- 44** **163** 38.7 254 **226**	**- 34** **161** 39.1 256 **228**	**- 24** **159** 39.7 260 **231**	**- 13** **158** 40.3 265 **233**	**- 3** **156** 40.9 267 **235**
24000	56.0	**- 46** **161** 37.9 249 **227**	**- 36** **159** 38.3 251 **230**	**- 25** **158** 38.8 254 **232**	**- 15** **156** 39.4 258 **234**	**- 5** **154** 40.0 262 **237**

Figure 5.10.23 (1/2) - CRUISE PERFORMANCE -
Long Range Cruise (5512 lbs - 2500 kg) (Altitude $\leq$ 24000 ft)

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Long Range Cruise (5512 lbs - 2500 kg) (Cont'd)

Conditions : Landing gear and flaps UP
2000 RPM (*) - BLEED ON

LEGEND : **IOAT**: °C **IAS** : KIAS
FF : us gal/h
FF : lbs/h **TAS**: KTAS

Altitude pression (feet)	TRQ (%)	ISA - 20°C	ISA - 10°C	ISA	ISA + 10°C	ISA + 20°C
24000	56.0	**- 46** **161** 37.9 249 **227**	**- 36** **159** 38.3 251 **230**	**- 25** **158** 38.8 254 **232**	**- 15** **156** 39.4 258 **234**	**- 5** **154** 40.0 262 **237**
25000	55.5	**- 48** **159** 37.2 243 **228**	**- 38** **158** 37.6 247 **231**	**- 27** **156** 38.1 249 **233**	**- 17** **154** 38.6 254 **236**	**- 7** **153** 39.2 256 **238**
26000	55.0	**- 50** **157** 36.5 238 **229**	**- 39** **156** 36.9 243 **232**	**- 29** **154** 37.4 245 **235**	**- 19** **152** 37.8 247 **237**	**- 9** **151** 38.3 251 **239**
27000	54.5	**- 52** **156** 35.8 234 **230**	**- 41** **154** 36.2 238 **233**	**- 31** **152** 36.7 240 **236**	**- 21** **151** 37.1 243 **238**	**- 11** **149** 37.6 247 **240**
28000	54.0	**- 53** **154** 35.1 229 **231**	**- 43** **152** 35.5 234 **235**	**- 33** **151** 36.1 236 **237**	**- 23** **149** 36.5 238 **239**	**- 13** **147** 37.0 243 **241**
29000	53.5	**- 55** **152** 34.6 227 **232**	**- 45** **150** 35.0 229 **236**	**- 35** **149** 35.5 231 **239**	**- 25** **147** 35.9 236 **241**	**- 15** **145** 36.3 238 **242**
30000	54.0	**- 57** **152** 34.4 225 **237**	**- 47** **150** 34.8 228 **239**	**- 37** **148** 35.2 231 **242**	**- 27** **146** 35.7 234 **244**	**- 17** **144** 36.1 236 **246**
31000	54.5	**- 59** **151** 34.2 224 **240**	**- 49** **149** 34.7 227 **243**	**- 39** **147** 35.1 230 **245**	**- 29** **145** 35.5 233 **247**	**- 19** **144** 35.9 235 **249**

Figure 5.10.23 (2/2) - CRUISE PERFORMANCE -
Long Range Cruise (5512 lbs - 2500 kg) (Altitude $\geq$ 24000 ft)

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Long Range Cruise (6173 lbs - 2800 kg)

Conditions : Landing gear and flaps UP
2000 RPM (*) - BLEED ON

LEGEND : **IOAT**: °C **IAS** : KIAS
FF : us gal/h
FF : lbs/h **TAS**: KTAS

Altitude pression (feet)	TRQ (%)	ISA - 20°C	ISA - 10°C	ISA	ISA + 10°C	ISA + 20°C
15000	65.5	**- 28** **179** 48.1 315 **218**	**- 18** **177** 48.8 320 **220**	**- 8** **176** 49.5 324 **223**	**+ 2** **174** 50.1 328 **225**	**+ 12** **172** 50.8 333 **227**
18000	64.0	**- 34** **174** 45.0 295 **222**	**- 24** **172** 45.7 300 **225**	**- 14** **171** 46.2 302 **227**	**- 4** **169** 47.0 309 **230**	**+ 6** **167** 47.6 311 **232**
19000	63.5	**- 36** **172** 44.1 289 **223**	**- 26** **170** 44.7 293 **226**	**- 16** **169** 45.3 298 **229**	**- 6** **167** 46.0 302 **231**	**+ 4** **166** 46.6 304 **233**
20000	63.0	**- 38** **170** 43.2 282 **225**	**- 28** **169** 43.7 287 **227**	**- 18** **167** 44.4 291 **230**	**- 8** **166** 45.0 295 **233**	**+ 3** **164** 45.6 298 **234**
21000	62.5	**- 40** **169** 42.3 278 **226**	**- 30** **167** 42.9 280 **229**	**- 20** **166** 43.5 284 **232**	**- 9** **164** 44.1 289 **234**	**+ 1** **162** 44.6 293 **236**
22000	62.0	**- 42** **167** 41.5 271 **227**	**- 32** **165** 42.0 276 **230**	**- 21** **164** 42.5 278 **233**	**- 11** **162** 43.2 282 **236**	**- 1** **161** 43.7 287 **238**
23000	61.5	**- 44** **165** 40.6 267 **229**	**- 34** **164** 41.1 269 **232**	**- 23** **162** 41.7 273 **235**	**- 13** **161** 42.3 278 **237**	**- 3** **159** 42.9 280 **239**
24000	61.0	**- 46** **164** 39.8 260 **230**	**- 35** **162** 40.3 265 **233**	**- 25** **161** 40.8 267 **236**	**- 15** **159** 41.5 271 **238**	**- 5** **157** 42.0 276 **240**

Figure 5.10.24 (1/2) - CRUISE PERFORMANCE -
Long Range Cruise (6173 lbs - 2800 kg) (Altitude $\leq$ 24000 ft)

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

CRUISE PERFORMANCE

Long Range Cruise (6173 lbs - 2800 kg) (Cont'd)

Conditions : Landing gear and flaps UP
2000 RPM (*) - BLEED ON

LEGEND : **IOAT**: °C **IAS** : KIAS
FF : us gal/h
FF : lbs/h **TAS**: KTAS

Altitude pression (feet)	TRQ (%)	ISA - 20°C		ISA - 10°C		ISA		ISA + 10°C		ISA + 20°C	
24000	61.0	**- 46** 39.8 260	**164** **230**	**- 35** 40.3 265	**162** **233**	**- 25** 40.8 267	**161** **236**	**- 15** 41.5 271	**159** **238**	**- 5** 42.0 276	**157** **240**
25000	60.5	**- 47** 39.0 256	**162** **232**	**- 37** 39.6 260	**160** **235**	**- 27** 40.1 262	**159** **237**	**- 17** 40.7 267	**157** **239**	**- 7** 41.2 269	**155** **241**
26000	60.0	**- 49** 38.3 251	**160** **233**	**- 39** 38.8 254	**159** **236**	**- 29** 39.4 258	**157** **239**	**- 19** 39.9 262	**155** **241**	**- 9** 40.4 265	**153** **243**
27000	59.5	**- 51** 37.6 247	**159** **235**	**- 41** 38.2 249	**157** **237**	**- 31** 38.7 254	**155** **240**	**- 21** 39.2 258	**153** **242**	**- 11** 39.8 260	**151** **244**
28000	59.0	**- 53** 37.0 243	**157** **236**	**- 43** 37.5 245	**155** **238**	**- 33** 38.0 249	**153** **241**	**- 23** 38.6 254	**151** **243**	**- 13** 39.1 256	**149** **245**
29000	58.5	**- 55** 36.5 238	**155** **238**	**- 45** 37.0 243	**153** **239**	**- 35** 37.5 245	**151** **242**	**- 25** 38.0 249	**149** **244**	**- 15** 38.5 251	**147** **246**
30000	59.0	**- 57** 36.4 238	**154** **240**	**- 47** 36.9 242	**152** **243**	**- 37** 37.3 244	**150** **245**	**- 27** 37.8 248	**148** **247**	**- 17** 38.2 250	**146** **248**
31000	59.5	**- 59** 36.3 238	**154** **244**	**- 49** 36.7 240	**152** **246**	**- 39** 37.2 244	**149** **248**	**- 29** 37.7 247	**147** **250**	**- 19** 38.1 250	**145** **252**

Figure 5.10.24 (2/2) - CRUISE PERFORMANCE -
Long Range Cruise (6173 lbs - 2800 kg) (Altitude $\geq$ 24000 ft)

(*) Propeller RPM utilization between 1600 and 2000 RPM is possible without changing performance. Display the TRQ indicated in table with Np = 2000 RPM, then reduce Np without resetting power lever (within limits permitted by torque limiter).

5.11 - TIME, CONSUMPTION AND DESCENT DISTANCE

Conditions : Power as required to maintain constant Vz
Landing gear and flaps UP
CAS = 230 KCAS - 2000 RPM - BLEED ON

Pressure altitude (feet)	Vz = 1500 ft/min					Vz = 2000 ft/min					Vz = 2500 ft/min				
	Time (min. s)	Consump.			Dist. (NM)	Time (min. s)	Consump.			Dist. (NM)	Time (min. s)	Consump.			Dist. (NM)
		l	kg	us gal			l	kg	us gal			l	kg	us gal	
30000	20.00	70	55	18.5	92	15.00	50	39	13.2	70	12.00	37	29	9.8	57
28000	18.40	67	53	17.7	85	14.00	47	37	12.4	65	11.10	34	27	9	52
26000	17.20	63	49	16.6	80	13.00	43	34	11.4	60	10.25	32	25	8.4	48
24000	16.00	58	45	15.3	72	12.00	41	32	10.8	55	09.35	29	23	7.7	43
22000	14.40	54	42	14.3	65	11.00	37	29	9.8	50	08.50	27	21	7.1	39
20000	13.20	49	39	12.9	58	10.00	34	27	9	45	08.00	24	19	6.3	35
18000	12.00	45	35	11.9	50	09.00	31	24	8.2	40	07.10	23	18	6.1	31
16000	10.40	40	31	10.6	45	08.00	28	22	7.4	35	06.25	20	16	5.3	27
14000	09.20	35	28	9.2	40	07.00	24	19	6.3	30	05.35	18	14	4.8	23
12000	08.00	31	24	8.2	33	06.00	20	16	5.3	25	04.50	15	12	4	20
10000	06.40	26	20	6.9	27	05.00	18	14	4.8	20	04.00	13	10	3.4	16
8000	05.20	21	16	5.5	20	04.00	14	11	3.7	16	03.10	10	8	2.6	13
6000	04.00	16	12	4.2	15	03.00	11	9	2.9	12	02.25	8	6	2.1	10
4000	02.40	10	8	2.6	10	02.00	8	6	2.1	8	01.35	5	4	1.3	6
2000	01.20	5	4	1.3	5	01.00	4	3	1.1	4	00.50	3	2	0.8	3
SL	00.00	0	0	0	0	00.00	0	0	0	0	00.00	0	0	0	0

Figure 5.11.1 - TIME, CONSUMPTION AND DESCENT DISTANCE

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5.12 - HOLDING TIME

Conditions : Landing gear and flaps UP
IAS = 120 KIAS - 2000 RPM - BLEED ON
TRQ ≃ 30 %

Pressure altitude (feet)	FUEL USED DURING HOLDING TIME											
	Weight 4850 lbs (2200 kg)						Weight 5512 lbs (2500 kg)					
	10 min			30 min			10 min			30 min		
	l	kg	us gal	l	kg	us gal	l	kg	us gal	l	kg	us gal
SL	29	23	7.7	87	69	23.0	31	24	8.2	93	72	24.6
5000	25	20	6.6	75	60	19.8	27	21	7.1	81	63	21.4
10000	23	18	6.1	69	54	18.2	24	19	6.3	72	57	19.0
15000	20	16	5.3	60	48	15.8	22	17	5.8	66	51	17.4
20000	19	15	5.0	57	45	15.0	20	16	5.3	60	48	15.8

Figure 5.12.1 - HOLDING TIME

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5.13 - LANDING DISTANCES

WEIGHT : 6250 lbs (2835 kg)

Associated conditions :
- Landing gear DN and flaps LDG
- Approach speed IAS = 80 KIAS
- Touch-down speed IAS = 65 KIAS
- Maximum braking without reverse
- Hard, dry and level runway
- GR = Ground roll (in ft)
- D_{50} = Landing distance (clear to 50 ft) (in ft)

PRESSURE ALTITUDE ft	ISA - 35°C		ISA - 20°C		ISA - 10°C		ISA	
	GR	D50	GR	D50	GR	D50	GR	D50
0	1050	1900	1115	2000	1180	2070	1215	2135
2000	1115	2000	1215	2100	1245	2200	1310	2265
4000	1180	2100	1280	2230	1345	2330	1410	2395
6000	1280	2230	1380	2360	1445	2460	1510	2525
8000	1380	2360	1475	2490	1540	2590	1610	2690
PRESSURE ALTITUDE ft	ISA + 10°C		ISA + 20°C		ISA + 30°C		ISA + 37°C	
	GR	D50	GR	D50	GR	D50	GR	D50
0	1280	2200	1310	2300	1380	2360	1445	2430
2000	1345	2330	1410	2430	1475	2495	1540	2560
4000	1445	2460	1510	2560	1575	2655	1640	2755
6000	1575	2645	1640	2720	1705	2820	1770	2920
8000	1705	2790	1770	2885	1835	2985	1900	3085

Figure 5.13.1 - LANDING DISTANCES - 6250 lbs (2835 kg)

Corrections :
. Reduce total distances of 10 % every 10 kt of headwind
. Increase total distances of 30 % every 10 kt of rear wind

Other runway surfaces require the following correction factors :

Increase by :
- 7 % on hard grass
- 10 % on short grass
- 15 % on wet runway
- 25 % on high grass
- 30 % on slippery runway

LANDING DISTANCES

WEIGHT : 5071 lbs (2300 kg)

Associated conditions :
- Landing gear DN and flaps LDG
- Approach speed IAS = 80 KIAS
- Touch-down speed IAS = 60 KIAS
- Maximum braking without reverse
- Hard, dry and level runway
- GR = Ground roll (in ft)
- D_{50} = Landing distance (clear to 50 ft) (in ft)

PRESSURE ALTITUDE ft	ISA - 35°C		ISA - 20°C		ISA - 10°C		ISA	
	GR	D50	GR	D50	GR	D50	GR	D50
0	885	1900	950	2000	1000	2070	1030	2135
2000	950	2000	1030	2100	1065	2200	1115	2265
4000	1000	2100	1080	2230	1150	2330	1200	2395
6000	1080	2230	1180	2360	1230	2460	1280	2525
8000	1180	2360	1245	2490	1310	2590	1360	2690
PRESSURE ALTITUDE ft	ISA + 10°C		ISA + 20°C		ISA + 30°C		ISA + 37°C	
	GR	D50	GR	D50	GR	D50	GR	D50
0	1080	2200	1115	2300	1180	2360	1230	2430
2000	1150	2330	1200	2430	1245	2495	1310	2560
4000	1230	2460	1280	2560	1345	2655	1395	2755
6000	1345	2645	1395	2720	1445	2820	1510	2920
8000	1445	2790	1510	2885	1560	2985	1610	3085

Figure 5.13.2 - LANDING DISTANCES - 5071 lbs (2300 kg)

Corrections :
. Reduce total distances of 10 % every 10 kt of headwind
. Increase total distances of 30 % every 10 kt of rear wind

Other runway surfaces require the following correction factors :

Increase by :
- 7 % on hard grass
- 10 % on short grass
- 15 % on wet runway
- 25 % on high grass
- 30 % on slippery runway